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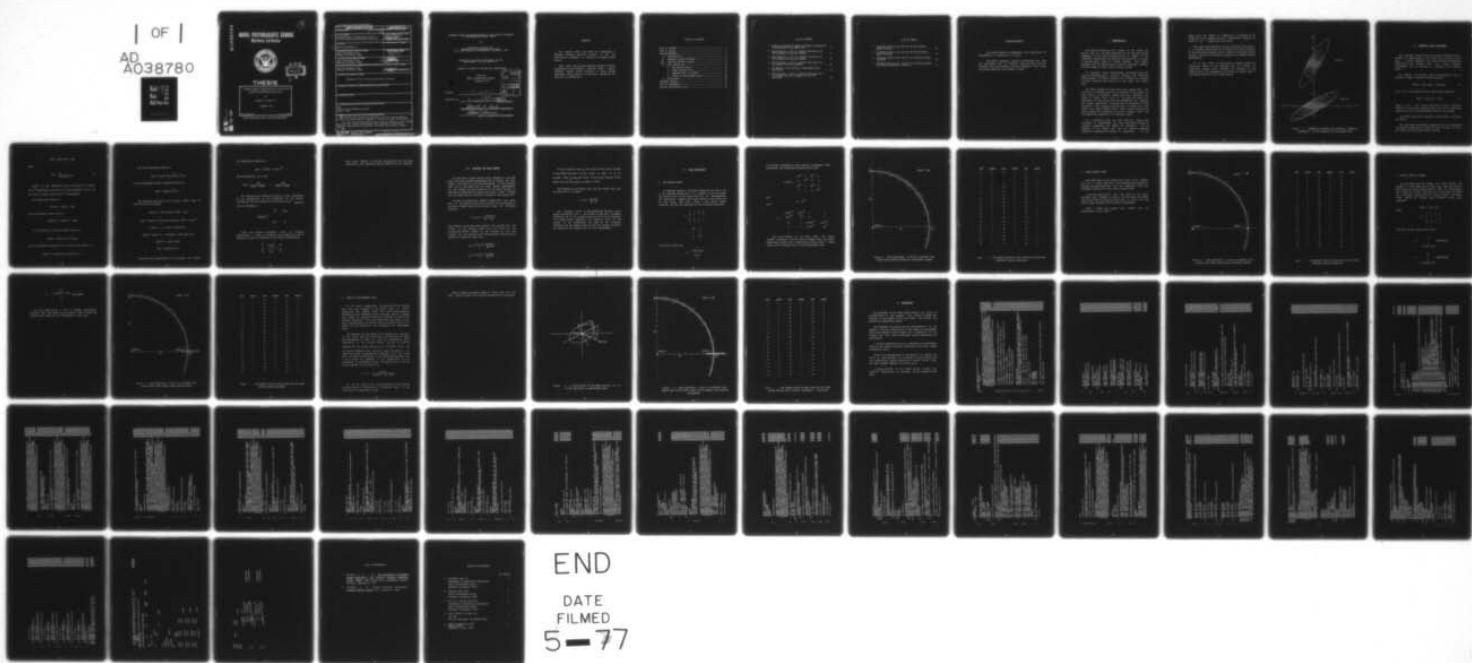
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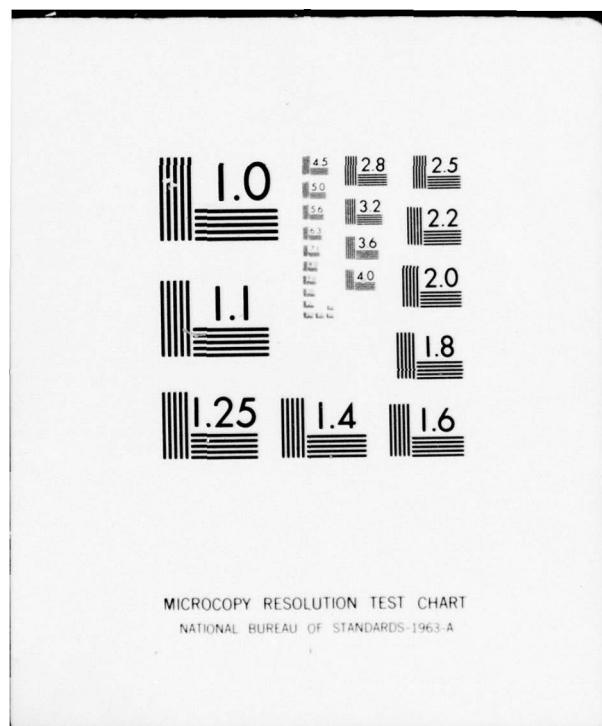
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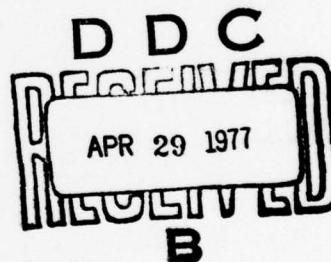


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Monterey, California



# THESIS

EXTENDED KALMAN FILTERING APPLIED TO THE POSITION  
LOCATING AND REPORTING SYSTEM (PLRS)

by

Bernard M. de Mahy, Jr.

December 1976

Thesis Advisor:

H. A. Titus

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This study has applied Extended Kalman Filtering techniques to that problem, evolving from a simple Extended Kalman Filter Observer to three moving observers, whose position is uncertain, estimating the position of another unit.			

EXTENDED KALMAN FILTERING APPLIED TO THE POSITION LOCATING  
AND REPORTING SYSTEM (PLRS)

by

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Submitted in partial fulfillment of the  
requirements for the degree of

MASTER OF SCIENCE IN ELECTRICAL ENGINEERING

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## ABSTRACT

The Marine Corps and Army are developing a Position Locating Reporting System to aid the battlefield commander in locating his assets during battle.

This study has applied Extended Kalman Filtering techniques to that problem, evolving from a simple Extended Kalman Filter Observer to three moving observers, whose position is uncertain, estimating the position of another unit.

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## I. INTRODUCTION

The precise location of all assets in and about the battle area is of prime importance to the tactical Marine Commander. In the past locating has had to depend on the individual knowing his own position and being able to report it through radio links to higher command. This system suffered from the limitations of terrain, daylight, weather, and the volume of radio traffic during battle.

To alleviate these shortcomings the Marine Corps and Army are investigating a Position Locating and Reporting System (PLRS) to collect, process, and, display the location of units, vehicles, and aircraft in and about the battle area.

The PLRS consists of field units and a master unit. The field unit is compact enough to be carried in the field by a man, vehicle, or aircraft. These units will determine the range to other field units in the area and report this information to the master unit for processing and display. The range information is determined by measuring the time required to send a signal from one unit to another and back again plus some "system" delay. When a unit's position is being updated it is referred to as the "Update" unit; and all others are referred to as "Ranging" units.

In a previous study in this area,[1], tests were conducted to investigate the use of the error ellipse in visually displaying the degree of uncertainty of the position of an update unit and the effect of numerous updates on reducing that degree of uncertainty. It was

found that the degree of uncertainty is reduced in the direction of the ranging unit with consecutive updates as shown in Fig 1 taken from that study.

That study also simulated one jet aircraft flying Mach 1 in a constant radius turn as an update unit being ranged on by two stationary ranging units to explore the proper random forcing excitation covariance necessary for adequate filter performance.

It is the intent of this study to further expand the simulation begun in the previous work by adding an additional ranging unit, allowing the movement of the ranging units, and considering the effect of ranging from a unit whose position is not known exactly

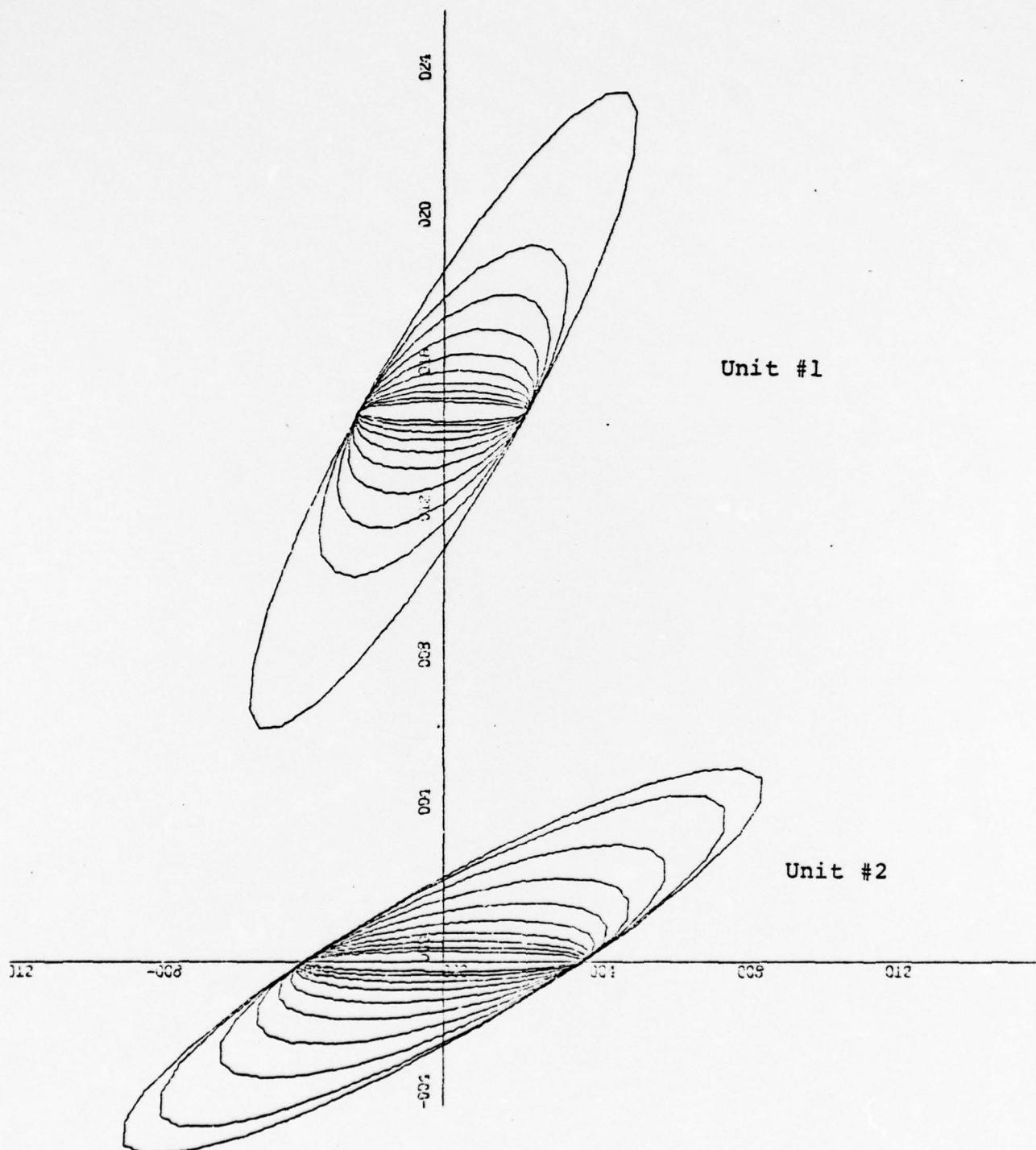


Figure 1 - CONSECUTIVE UPDATES WILL REDUCE THE DEGREE OF  
UNCERTAINTY IN THE DIRECTION OF THE RANGING UNIT

### III. EXTENDED KALMAN FILTERING

The Extended Kalman Filter is widely documented and no attempt at a development of that theory will be made in this work. A brief treatment has been included to establish nomenclature and formulas used. For a more complete development one is referred to reference [2] or similar texts.

As defined in this work, PLRS is described by a set of discrete, linear, cartesian system equations

$$\underline{x}(k+1) = \underline{\phi}(k) \underline{x}(k) + \underline{\Gamma}(k) \underline{w}(k) \quad (1)$$

and a set of discrete non-linear measurement equations

$$\underline{z}(k) = \underline{m}(\underline{x}(k), k) + \underline{v}(k) \quad (2)$$

where  $\underline{\phi}$  and  $\underline{\Gamma}$  are linear functions and  $\underline{m}$  is a nonlinear function of the state variables  $\underline{x}(k)$ ;  $\underline{w}(k)$  is the excitation noise and  $\underline{v}(k)$  is the measurement noise of the system.

The plant noises are considered uncorrelated, zero-mean, and white.

The non-linear measurement equations can be linearized by expanding equation (2) around the best estimate at time  $k$  and using the first-order terms yielding

$$\underline{z}(k) = \underline{H}(k) \underline{x}(k) + \underline{v}(k)$$

where

$$\underline{H}(k) = \frac{\partial \underline{m}}{\partial \underline{x}} \underline{x} = \hat{\underline{x}}(k/k-1) \quad (3)$$

$\hat{\underline{x}}(k/k)$  is the estimated value of the state at  $k$  after the  $k^{\text{th}}$  measurement and  $\hat{\underline{x}}(k/k-1)$  is the predicted value of the state at time  $k$  before the  $k^{\text{th}}$  measurement.

The state error vector is

$$\underline{x}^*(k/k) = \hat{\underline{x}}(k/k) - \underline{x}(k)$$

and the predicted error vector is

$$\hat{\underline{x}}^*(k/k-1) = \hat{\underline{x}}(k/k-1) - \underline{x}(k)$$

The covariance of the state error matrix is

$$P(k/k) = E[\hat{\underline{x}}^*(k/k) \hat{\underline{x}}^{*T}(k/k)]$$

and the predicted covariance of the state error matrix is

$$P(k/k-1) = E[\hat{\underline{x}}^*(k/k-1) \hat{\underline{x}}^{*T}(k/k-1)] .$$

The state excitation matrix is

$$Q(k) = E[\underline{L}(k) \underline{u}(k) \underline{u}^T(k) \underline{L}^T(k)]$$

and the measurement noise covariance matrix is

$$R(k) = E[\underline{v}(k) \underline{v}^T(k)] .$$

The equations that made up the Kalman Filter used in this work are as follows:

$$P(k/k-1) = \underline{\Phi}(k) P(k/k) \underline{\Phi}^T(k) + Q(k)$$

$$G(k) = P(k/k-1) \underline{H}^T(k) [ \underline{H}(k) P(k/k-1) \underline{H}^T(k) + R(k) ]^{-1}$$

$$P(k/k) = [I - G(k) \underline{H}(k)] P(k/k-1)$$

$$\hat{\underline{x}}(k/k) = \hat{\underline{x}}(k/k-1) + G(k) [\underline{z}(k) - \underline{H}(k) \hat{\underline{x}}(k/k-1)]$$

$$\hat{\underline{x}}(k/k-1) = \underline{\Phi}(k) \hat{\underline{x}}(k/k)$$

$$\underline{z}(k) = \underline{m}(\underline{x}(k/k-1), k)$$

Since the only observations in this system are ranges,

the observation equation is

$$\underline{z}(k) = [x^2(k) + y^2(k)]^{1/2};$$

and from equation (3) we get

$$H(k) = \begin{bmatrix} \frac{x(k)}{x^2(k) + y^2(k)} & 0 \\ 0 & \frac{y(k)}{x^2(k) + y^2(k)} \end{bmatrix}.$$

The covariance of estimation error,  $P$ , is an expression of the uncertainty in the estimation of the states. Considering only the estimation's position error,  $P_{\text{position}}$  can be expressed as

$$P_{\text{position}} = \sigma_x^2 \quad \sigma_x \sigma_y$$

$$P_{\text{position}} = \frac{\sigma_y \sigma_x}{\sigma_y^2} \quad \sigma_y^2$$

Since the position estimation error is normally distributed, a curve of constant error probability can be defined by using the exponent of the normal distribution,

$$\frac{x^2}{\sigma_x^2} - \frac{2r_{xy}}{\sigma_x \sigma_y} + \frac{y^2}{\sigma_y^2}$$

This curve defines an ellipse. Graphically, for the given probability, the estimation may be anywhere in that ellipse.

### III. CHOOSING THE BEST RANGER

To move from a simple Kalman Filter observer to the PLRS model the first problem encountered was to choose the best ranger from which to take the measurement. In the previous work , [1], it was shown that the most useful measurement, the one causing the most reduction in the error ellipse, is obtained by observing the update unit from a point aligned with the major axis of the error ellipse of the update unit.

To find the ranger most closely aligned with the major axis of the update unit's error ellipse the orientation of the error ellipse must first be found using the following equation.

$$\theta = \frac{1}{2} \tan^{-1} \frac{2 \operatorname{Cov}(x, y)}{\sigma_x^2 - \sigma_y^2}$$

This angle( $\theta$ ) gives the angle between  $-90^\circ$  and  $90^\circ$  that the x-axis of the ellipse makes with the x-axis of the co-ordinate system. Looking at the ellipse in this new posture one can find the new "Uncorrelated" variances that define the major and minor axes.

$$\sigma_x^{2*} = \frac{\sigma_x^2 + \sigma_y^2}{2} + \frac{\operatorname{Cov}(x, y)}{\sin 2\theta} ,$$

$$\sigma_y^{2*} = \frac{\sigma_x^2 + \sigma_y^2}{2} - \frac{\operatorname{Cov}(x, y)}{\sin 2\theta} ,$$

If  $\sigma_x^2$  is greater than  $\sigma_y^2$  the x-axis of the error ellipse is the major axis and  $\theta$  is the angle we seek. If  $\sigma_y^2$  is greater than  $\sigma_x^2$  then the y-axis of the error ellipse is the major axis and the angle we seek is  $\theta + 90^\circ$ .

The bearing of the update unit from the ranger must then be found and it is simply

$$\beta = \tan^{-1} \frac{y_u - y_R}{x_u - x_R}.$$

The absolute value of the difference between  $\theta$ , after proper correction, and  $\beta$  was chosen as the best alignment indicator; but to be aligned and to be  $180^\circ$  out of alignment is of equal value; therefore the absolute value of the cosine of the differences was used as the alignment indicator and the ranger found to have the largest indicator was chosen as the ranging unit for that measurement.

#### IV. PLRS SIMULATION

##### A. TWO RANGING UNITS

In previous work,[1], the PLRS simulation was setup for a jet aircraft flying Mach 1 in a constant 10 Km turn about the origin to act as the update unit for all measurements. Two stationary ranging units were placed at the origin and at 10Km north, 10Km east. Using a one second sample interval, the jet was described by the following matrices:

$$\phi = \begin{matrix} 1 & 1 & 0 & 0 \\ 0 & 1 & 0 & 0 \\ 0 & 0 & 1 & 1 \\ 0 & 0 & 0 & 1 \end{matrix}$$

$$\Gamma = \begin{matrix} 0.5 & 0 \\ 1 & 0 \\ 0 & 0.5 \\ 0 & 1 \end{matrix}$$

Its initial state was

$$\underline{x} = \begin{matrix} 0 \\ 0.333 \text{ Km/s} \\ 10 \text{ Km} \\ 0 \end{matrix}$$

Its initial covariance of error matrix, measurement noise covariance, and excitation forcing matrix were

$$P(1/0) = \begin{matrix} 10^{-4} & 0 & 10^{-4} & 0 \\ 0 & 10^{-4} & 0 & 0 \\ 10^{-4} & 0 & 10^{-4} & 0 \\ 0 & 0 & 0 & 10^{-4} \end{matrix}$$

and

$$R = 10^{-4}$$

with

$$Q = \begin{matrix} 2.5 \times 10^{-5} & 5 \times 10^{-5} & 0 & 0 \\ 5 \times 10^{-5} & 10^{-4} & 0 & 0 \\ 0 & 0 & 2.5 \times 10^{-5} & 5 \times 10^{-5} \\ 0 & 0 & 5 \times 10^{-5} & 10^{-4} \end{matrix}$$

Fig 2 is a display of its final runs. The filter tracked accurately and the error ellipses shown are twenty times their actual size to make them visible. Table 1 shows which was the ranging unit at each measurement time.

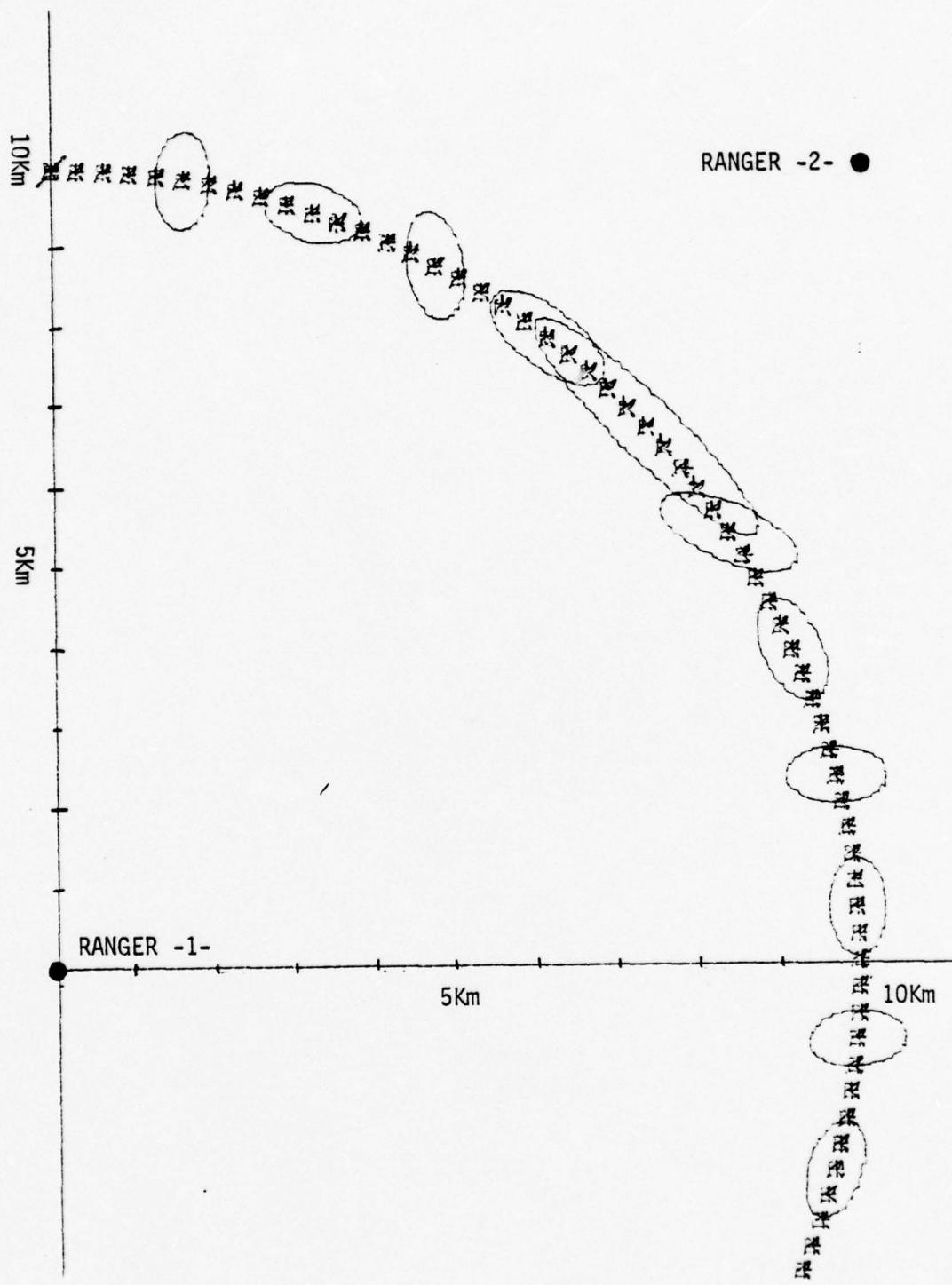


Figure 2 - PLRS SIMULATION - A JET IN A CONSTANT 10KM  
RADIUS TURN FLYING BETWEEN TWO STATIONARY RANGERS

TIME	RANGER	TIME	RANGER	TIME	RANGER
1	2	21	2	41	1
2	1	22	1	42	2
3	2	23	2	43	1
4	1	24	1	44	2
5	2	25	1	45	1
6	1	26	2	46	2
7	2	27	1	47	1
8	1	28	2	48	2
9	2	29	1	49	1
10	1	30	2	50	2
11	2	31	1	51	1
12	1	32	2	52	2
13	2	33	1	53	1
14	1	34	2	54	2
15	2	35	1	55	1
16	1	36	2	56	2
17	2	37	1	57	1
18	1	38	2	58	2
19	2	39	1	59	1
20	1	40	2	60	2

TABLE 1 - THE RANGER CHOSEN AT EACH TIME FOR THE PLRS TWO STATIONARY RANGER SIMULATION

## B. THREE RANGING UNITS

The first step of this study was to add a third ranging unit at 0 north, 10Km east. The algorithm was enlarged to include the additional unit and its comparison with the alignment indicators of the other ranging units.

It can be seen in Fig 3 that the size of the error ellipses were reduced in size in the mid-range area where the jet and the two original units were in line; and the third ranger provides the triangular measurement.

Table 2 shows the ranging unit chosen for the measurement at each time k.

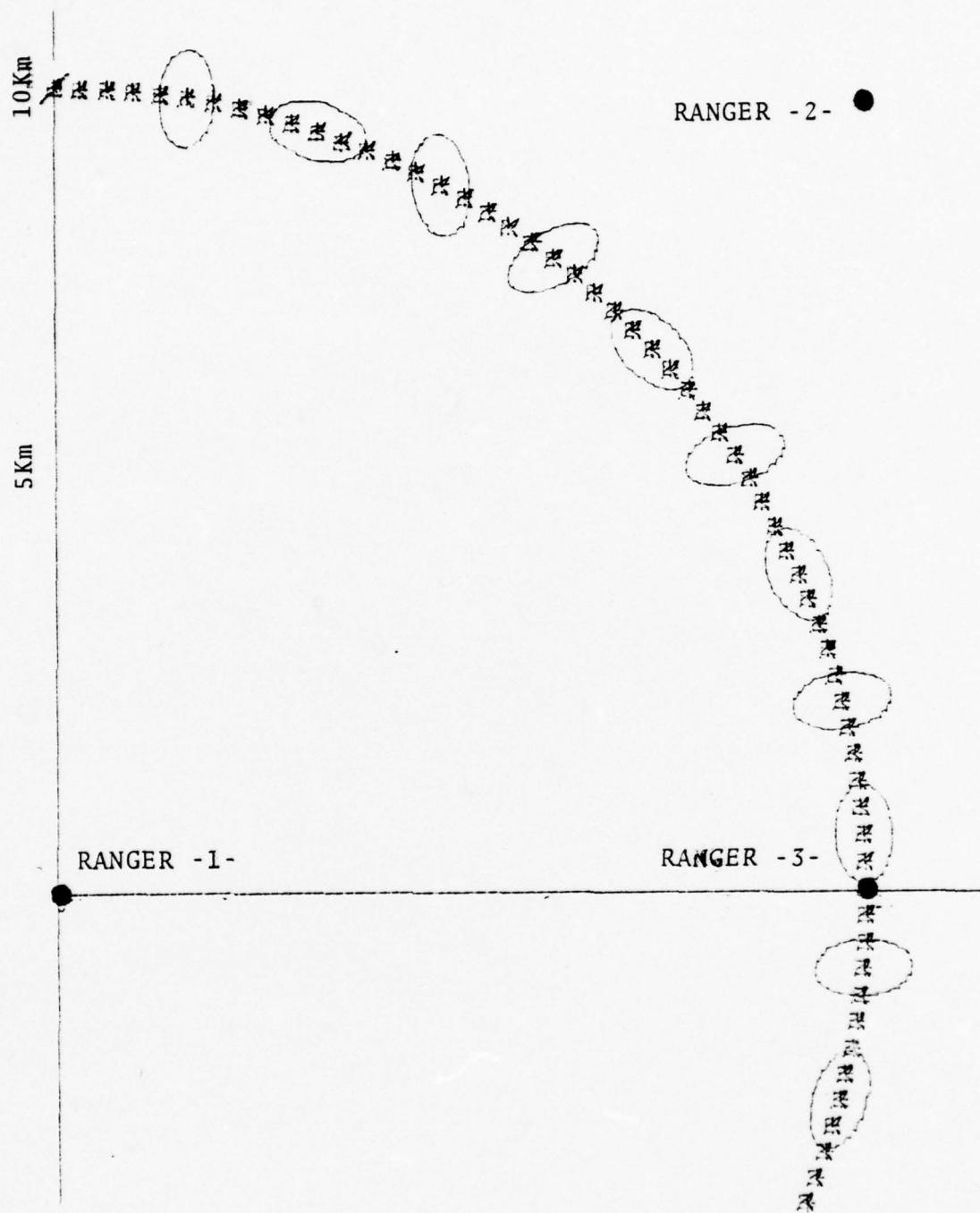


Figure 3 - PLRS SIMULATION - A JET IN A CONSTANT 10KM  
RADIUS TURN FLYING AMONG THREE STATIONARY RANGERS

TIME	RANGER	TIME	RANGER	TIME	RANGER
1	2	21	2	41	1
2	1	22	3	42	3
3	2	23	2	43	1
4	1	24	3	44	3
5	2	25	1	45	1
6	1	26	3	46	3
7	2	27	1	47	1
8	1	28	3	48	2
9	2	29	1	49	1
10	1	30	3	50	2
11	2	31	1	51	1
12	1	32	3	52	3
13	2	33	1	53	1
14	3	34	3	54	3
15	2	35	1	55	1
16	3	36	3	56	3
17	2	37	1	57	1
18	3	38	3	58	3
19	2	39	1	59	1
20	3	40	3	60	3

TABLE 2 - THE RANGER CHOSEN AT EACH TIME FOR THE THREE STATIONARY RANGER SIMULATION

### C. RANGING UNITS IN MOTION

In the second step the rangers are given motion. The rangers at the origin and at 10Km north, 10Km east were to move north and south respectively at 3Kts as infantrymen. The ranger at 0 north, 10Km east was to move west at 120Kts as a helicopter. Again using one second sample intervals, their motion was defined using discrete linear state equations

$$x(k+1) = \phi(k) x(k) ,$$

where

$$\phi = \begin{matrix} 1 & 1 & 0 & 0 \\ 0 & 1 & 0 & 0 \\ 0 & 0 & 1 & 1 \\ 0 & 0 & 0 & 1 \end{matrix}$$

with the initial states shown below;

$$x = \begin{matrix} 0 \\ 0 \\ 0 \\ 1.67 \times 10^3 \text{ Km/s} \end{matrix} \text{ INFANTRYMAN}$$

$$x = \begin{matrix} 10 \\ 0 \\ 10 \\ -1.67 \times 10^3 \text{ Km/s} \end{matrix} \text{ INFANTRYMAN}$$

12

$x = -5.555 \times 10^2$  Km/s      HELICOPTER

0

0

It can be seen in Fig 4 that no system depreciation resulted from the motion of the rangers, Table 3 shows the ranging unit chosen for the measurement at each time.

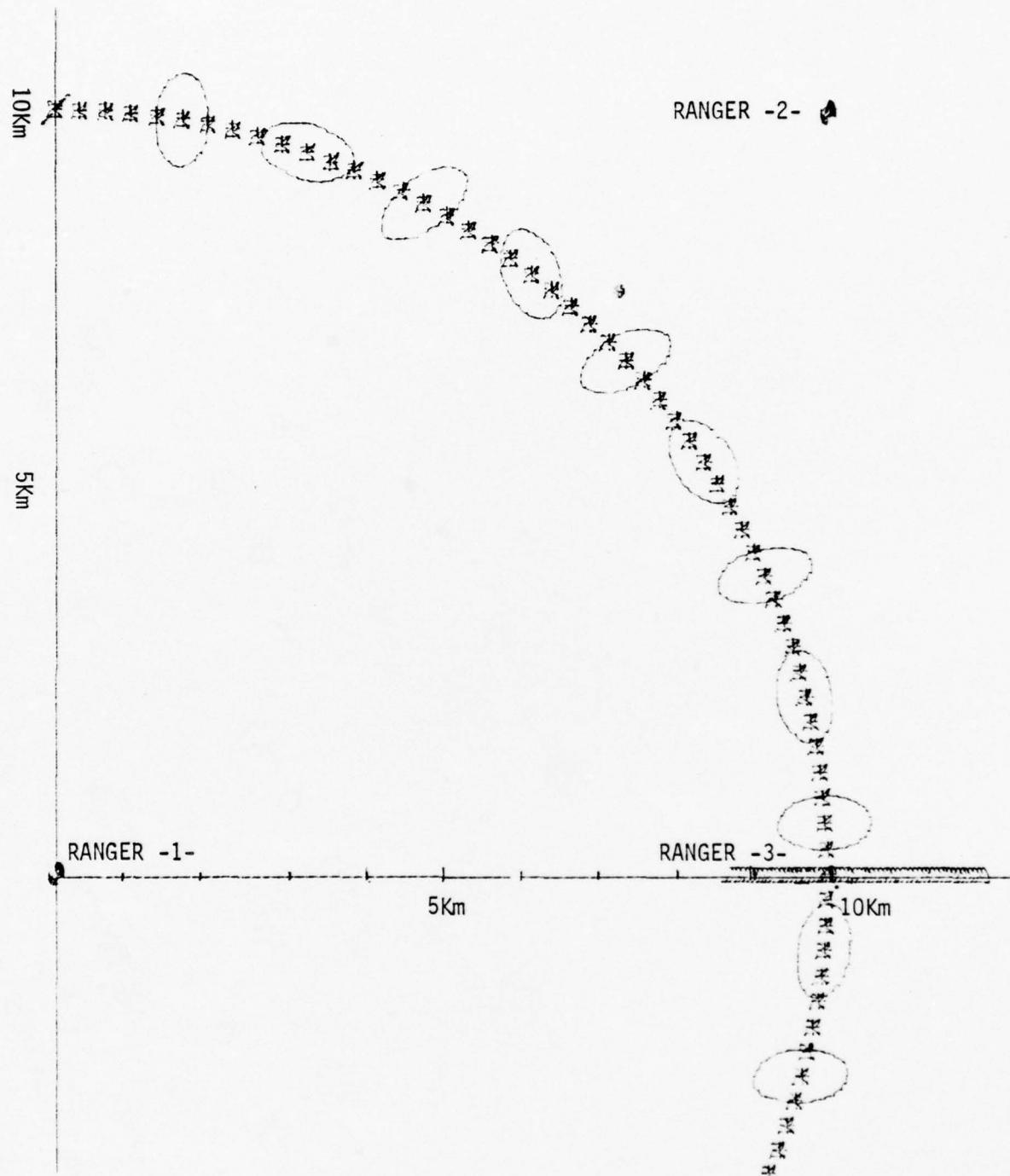


Figure 4 - PLRS SIMULATION - A JET IN A CONSTANT 10 KM RADIUS TURN FLYING AMONG THREE MOVING RANGERS

TIME	RANGER	TIME	RANGER	TIME	RANGER
1	2	21	3	41	3
2	1	22	2	42	1
3	2	23	3	43	2
4	1	24	2	44	1
5	2	25	3	45	2
6	1	26	1	46	1
7	2	27	3	47	2
8	1	28	1	48	1
9	2	29	3	49	2
10	1	30	1	50	1
11	2	31	3	51	2
12	1	32	1	52	1
13	3	33	3	53	2
14	1	34	1	54	1
15	3	35	3	55	2
16	2	36	1	56	1
17	3	37	3	57	2
18	2	38	1	58	1
19	3	39	3	59	2
20	2	40	1	60	1

TABLE 3 - THE RANGER CHOSEN AT EACH TIME FOR THE THREE MOVING RANGER SIMULATION

#### D. SOURCE OF MEASUREMENT NOISE

In the above simulations the position of the ranging unit has been assumed to be exact; while in actual application the ranging units will have covariances of estimation error defining an error ellipse; and the ranging unit might be anywhere within that ellipse. To bring this position uncertainty into the simulation, the radius of the error ellipse along the bearing from the ranging unit to the update unit was defined as the covariance of measurement error.

The equation for the radius of an ellipse is a function of the major axis, the minor axis, and the angle at which the measurement is made. To find the measurement noise covariance, or the ellipse radius,  $\sigma_x^2$  and  $\sigma_y^2$  must be compared and the larger defined as  $M_j$ , the major axis, and the smaller defined as  $M_n$ , the minor axis. The angle,  $\alpha$ , at which the radius is determined is measured from the major axis and thus is calculated as the difference between  $\theta$  and  $\beta$ . Fig 5 shows the geometry of the calculation of the covariance of measurement noise. The equation for  $R$  and the radius squared of the ellipse is:

$$R = r^2 = \frac{M_j M_n}{M_j \sin^2 \alpha + M_n \cos^2 \alpha}$$

It can be seen in Fig 6 that performance was improved slightly using the covariance of estimation error as the sole source of measurement noise.

Table 4 shows the ranger chosen at each time for the three moving rangers with position uncertainty simulation.

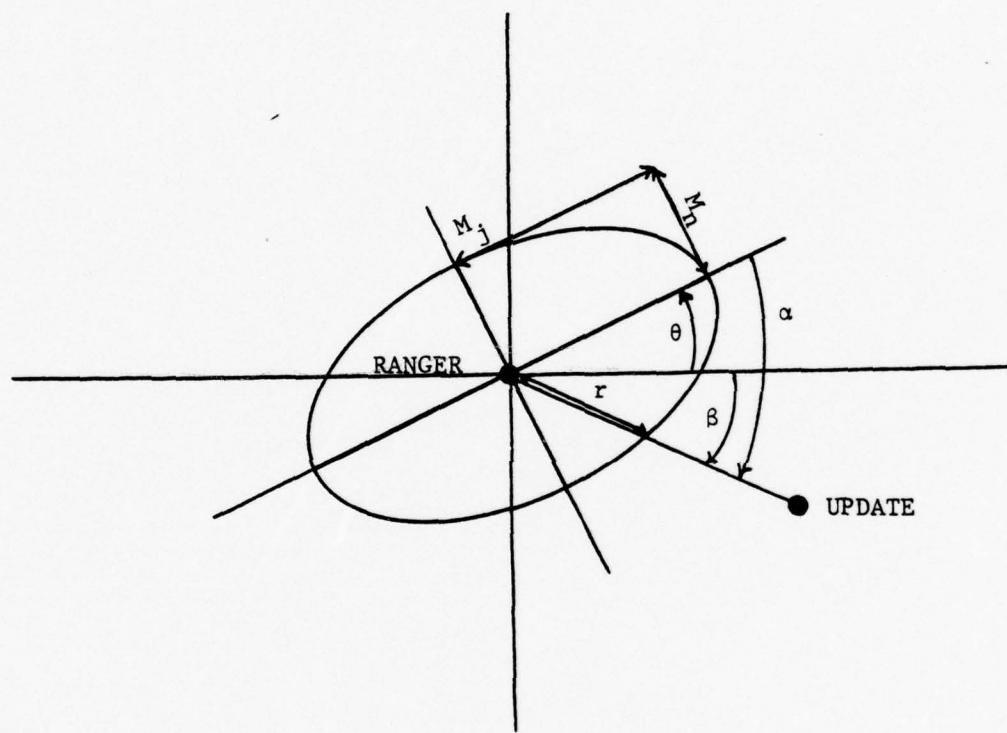


Figure 5 -  $r$  IS THE RADIUS OF THE ERROR ELLIPSE -  $r^2 = R$   
IS THE COVARIANCE OF MEASUREMENT NOISE

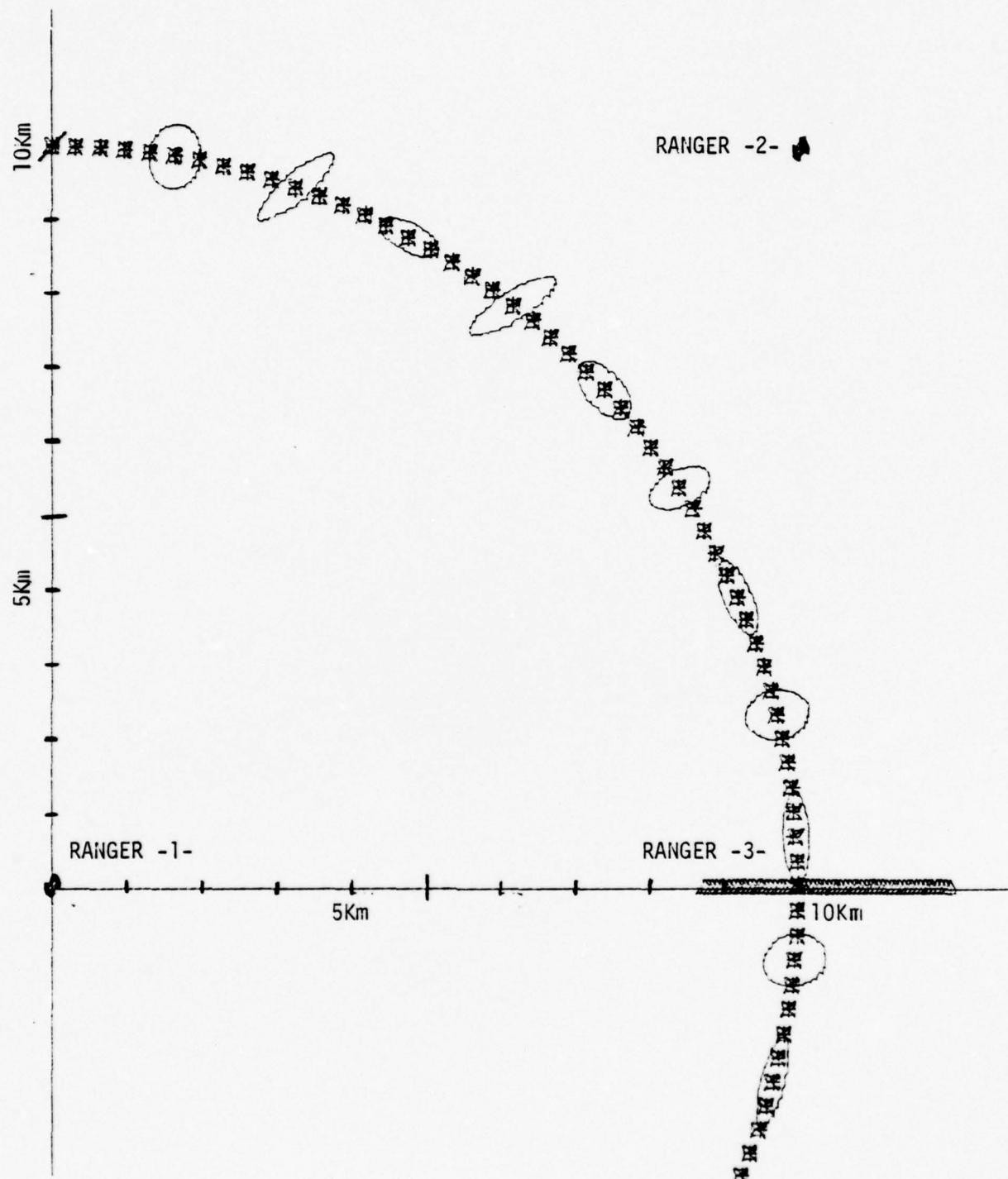


Figure 6 - PLRS SIMULATION - A JET IN A CONSTANT 10KM RADIUS TURN FLYING AMONG THREE MOVING RANGERS WITH POSITION UNCERTAINTY

TIME	RANGER	TIME	RANGER	TIME	RANGER
1	2	21	2	41	1
2	1	22	3	42	2
3	2	23	2	43	1
4	1	24	3	44	2
5	2	25	1	45	1
6	1	26	3	46	2
7	2	27	1	47	1
8	1	28	3	48	2
9	2	29	1	49	1
10	3	30	3	50	2
11	1	31	1	51	1
12	3	32	3	52	2
13	1	33	1	53	1
14	3	34	3	54	2
15	1	35	1	55	1
16	3	36	3	56	2
17	1	37	1	57	1
18	3	38	3	58	2
19	2	39	1	59	1
20	3	40	3	60	2

TABLE 4 - THE RANGER CHOSEN AT EACH TIME FOR THE THREE MOVING RANGERS WITH POSITION UNCERTAINTY SIMULATION

## V. CONCLUSION

The placement of the third ranger showed the value of triangulation of the rangers. The closer to normal the bearings of the rangers are to each other, the better the results of consecutive ranges.

The allowance for motion and the representation of the ranger's position uncertainty as the source of measurement error were important steps toward full simulation of the system; and they were accomplished without degradation of performance.

A better simulation may be to represent the measurement error as the ranger's position uncertainty plus some system measurement error.

Still to be accomplished is the ability to update all units at each ranging, and to provide a gating system that will demand more frequent updates for faster moving units and less frequent updates for slower units.

A program listing of the three moving rangers with position uncertainty is included with an annotated data deck.



```

C      DC 23 I=1,N
      DO 23 J=1,N      PHI(I,J)
23    PHI(I,J)=PHI(I,J)
      WRITE(6,131)
      CALL MWRITE(PHI,N,N)

C      CALL MREAD(H,M,N)
      DO 25 I=1,M
      DO 25 J=1,N      H(I,J)=H(I,J)
25    H(I,J)=H(I,J)
      WRITE(6,132)
      CALL MWRITE(H,M,N)

C      CALL MREAD(R,M,M)
      WRITE(6,133)
      CALL MWRITE(R,M,M)

C      CALL MREAD(COVW,IN,IN)
      WRITE(6,134)
      CALL MWRITE(COVW,IN,IN)
      CALL MREAD(GAMMA,N,IN)

C      DO 30 I=1,N
      DO 30 J=1,IN      GAMMAS(I,J)=GAMMA(I,J)
30    WRITE(6,136)
      CALL MWRITE(GAMMA,N,IN)

C      CALL MREAD(PKKM1,N,N)
      WRITE(6,137)
      CALL MWRITE(PKKM1,N,N)

C      DO 311 K=2,NR
      CALL MREAD(PRR,N,N)
      DO 310 I=1,N
      DO 310 J=1,N      PR(I,J,K)=PRR(I,J)
310  PR(I,J,K)=PRR(I,J)
311  CCNTINUE

C      CALL VREAD(SIGV,M)
      WRITE(6,138)

```

```

C CALL VWRITE (SIGV,M)
C
C DO 340 I=1, NR
C   READ (5,144) (XHATZ(I,J), J=1,N)
C   WRITE (6,140) (XHATZ(I,J), J=1,N)
C
C 36 DO 360 I=1, NR
C   READ (5,144) (XS(I,J,1), J=1,N)
C   INITIAL CONDITION HAS BEEN READ
C   WRITE (6,143) (XS(I,J,1), J=1,N)
C 360 WRITE (6,146)(XS(I,J,1), J=1,N)
C
C 38 CALL TRACK
C
C DO 390 K=1, NR
C   WRITE (6,145) (XS(K,I,1), I=1,N)
C   WRITE (6,146) (XS(K,I,NSAM), I=1,N)
C 390 CONTINUE
C
C THE FOLLOWING SECTION PREPARES FOR THE MONTE CARLO LOOP
C FORM NXN IDENTITY MATRIX IN DOUBLE PRECISION
C
C DO 41 I=1, N
C   DO 41 J=1, N
C     EI(I,J)=0. DO
C 41 IF (I.EQ.J) EI(I,J)=1. DO
C
C GIVING THE MATRIX GAMMA AND THE COVARIANCE OF W COMPUTE Q
C USING DOUBLE PRECISION ARITHMETIC
C
C CALL QMAT
C   WRITE (6,135)
C   CALL MWRITE (Q,N,N)
C
C SET UP ARRAYS FOR COMPUTING STATISTICS
C
C DO 48 I=1, NR
C   DO 48 K=1, NSAM
C     DC 48 J=1, N

```

```

XM(I,J,K) = 0.
ERR(I,J,K) = 0.

C   DO 48 L=1,N
C   VAR(J,L,K) = 0.

C   BEGIN MAIN ITERATION LOOP HERE
DC 54 ITER=1,NENS

C   DO 50 I=1,N
50 XHKKM1(I) = XHATZ(I,I)

DO 54 K=1,NSAM
FORM NOISY MEASUREMENT FROM TRUE STATE VALUE

C   DO 51 I=1,N
51 X(I) = X${I,I,K}
CALL GAIN

C   DO 52 I=1,N
DO 52 J=1,M
52 GKS(I,J,K) = G(I,J)

C   UPDATE THE STATE ESTIMATE
53 CALL ESTIM

C   UPDATE RUNNING SUMS USED IN COMPUTING STATISTICS
CALL STAT

C   CONTINUE

C   DIVIDE RUNNING SUMS COMPUTED BY SUBROUTINE STAT BY ENSEMBLE
SIZE TO COMPUTE STATISTICS
ENS = NENS

DO 56 K=1,NSAM
DO 56 J=1,N
55 ERR(I,J,K) = ERR(I,J,K)/ENS

```





THE APPROPRIATE STATEMENTS FOR COMPUTING R ON-LINE MUST  
BE INSERTED HERE BY THE USER

```

      RETURN
      END

      SUBROUTINE STAT
      THIS SUBROUTINE COMPUTES RUNNING SUMS USED IN DETERMINING THE
      SAMPLE STATISTICS OF TRACK AND ESTIMATION ERRORS. IN THE DEFAULT
      OPTION (ISTAT.EQ.0) THE STATISTICS TO BE COMPUTED ARE MEAN OF
      TRACK, MEAN OF ESTIMATION ERROR AND VARIANCE OF ESTIMATION
      ERROR. IF (ISTAT.NE.0) THE OFF-DIAGONAL TERMS IN THE COVARIANCE OF
      ESTIMATION ERROR MATRIX ARE ALSO COMPUTED. MCSP0832
      REAL*8 GAMMA COVW R2 PHI H TEMP1 TEMP2 PKK Q4 PKK Q4 PR
      COMMON EI(4,4)1 Q(4,4)1 G(4,4)1 GAMMA(4,4)1 COVW(4,4)1
      TEMP1(4,4)1 TEMP2(4,4)1 PKK(4,4)1 H(4,4)1 PHI(4,4)1 R(4,4)1
      2 VAR(4,4)1 GKS(4,4)1 PPKS(4,4)1 XM(4,4)1 ERR(4,4)1
      3 GAMMAS(4,4)1 PHYS(4,4)1 XS(4,4)1 HS(4,4)1 SIGW(4,4)1
      4 SIGX2(4,4)1 XZMEAN(4,4)1 XHKK(4,4)1 VTMP(4,4)1 X(4,4)1
      5 XHATZ(4,4)1 XZ(60)1 YZ(60)1 PY(10)1 PR(4,4)1
      6 NINSAM1 IQ M ITER ITRK IN ISTAT K ITRD IXZ IV IW IEST ND NR
      DIMENSION EXH(3)1
      IF (ITRK.NE.1) GO TO 2
      IF (ITER.NE.1) GO TO 4
      DO 1 J=1,N = XS(1,J,K)
      1 XM(1,J,K) = XS(1,J,K)
      GO TO 4
      2 CONTINUE
      3 DC 3 J=1,N = XM(1,J,K)+XS(1,J,K)
      4 CONTINUE
      5 DC 5 J=1,N
      EXH(J) = XHKK(J)-XS(J,J,K)
      ERR(1,J,K) = ERR(1,J,K)+EXH(J)
      VAR(J,J,K) = VAR(J,J,K)+EXH(J)*2
      IF (ISTAT.EQ.0) RETURN
      DC 6 L=2,N
      LM1 = L-1

```

```

      DO 6 J=1,LM1 = VAR(L,J,K)+EXH(L)*EXH(J)
      C
      C      RETURN
      C
      C      SUBROUTINE XZERO
      C      THIS SUBROUTINE GENERATES THE INITIAL STATE VALUE FROM A NORMAL
      C      RANDOM NUMBER GENERATOR. IT IS ASSUMED THAT THE INITIAL STATE
      C      HAS COMPONENTS THAT ARE INDEPENDENT
      C      REAL*8 GAMMA, COVM, RH, TEMP, TEMP1, TEMP2, PKK, QE1, PR
      C      COMMON E1(4,4), Q(4,4), G(4,4), PKK(4,4), GAMMA(4,4), COVM(4,4)
      C      TEMP(4,4), TEMP1(4,4), TEMP2(4,4), H(4,4), PHI(4,4), R(4,4)
      C      2VAR(4,4,60), PKKS(4,4,60), X(4,4,60), X(4,4,60), ERR(4,4,60)
      C      3GAMMA(4,4), PHIS(4,4), X(4,4,60), HS(4,4,60), GK(4,4,60)
      C      4SIGXZ(4,4), XZMEAN(4), XHKK(4), XHKKM(4), VTMP(4,2,4), SIGN(4,2,4)
      C      5XHATZ(4,4), XZ(60), YZ(60), PX(10), PY(10), PR(4,4,4)
      C      6NNSAM, IQ, MITER, ITRK, IN, ISTAT, K, ITRO, IXZ, IV, IW, IEST, ND, NR
      C      CALL SNORM(IXZ, X, N)
      C
      C      DO 1 I=1,N
      C      1 XS(I,I,I) = SIGXZ(I)*X(I)+XZMEAN(I)
      C
      C      RETURN
      C
      C      SUBROUTINE ADD(A,B,NM,C)
      C      THIS SUBROUTINE ADDS THE NXM MATRICES A AND B, STORING THE
      C      RESULT IN C
      C      REAL*8 A, B, C
      C      DIMENSION A(4,4), B(4,4), C(4,4)
      C
      C      DO 1 I=1,N
      C      DO 1 J=1,M
      C      1 C(I,J) = A(I,J)+B(I,J)
      C
      C      RETURN
      C
      C      SUBROUTINE MREAD(A,N,M)
      C      8010.5 THE ENTRIES IN THE FIRST ROW OF A ARE READ FIRST, THEN
      C      THIS SUBROUTINE READS AN NXM MATRIX A ACCORDING TO THE FORMAT
      C      THE ENTRIES IN THE SECOND ROW, AND SO ON.
      C      REAL*8 A
      C      DIMENSION A(4,4)
      C
      C      DO 1 I=1,N
      C      1 READ(5,2) (A(I,J), J=1, M)
      C
      C      RETURN

```

```

C 2 FORMAT (8F10.0)
C END
C SUBROUTINE MMWRITE (A,N,M)
C THIS SUBROUTINE WRITES THE ENTRIES OF THE NXM MATRIX A
C REAL*8 A
C DIMENSION A(4,4)
C
C DC 1 I=1,N
C 1 WRITE (6,2) (A(I,J),J=1,M)
C
C RETURN
C
C 2 FORMAT (9(2X,1PE12.5))
C
C END
C SUBROUTINE PROD (A,B,N,M,L,C)
C THIS SUBROUTINE COMPUTES THE MATRIX PRODUCT AB AND STORES THE
C RESULT IN C
C A = NXM A, B = MXL, C = NXL
C REAL*8 A, B, C
C DIMENSION A(4,4), B(4,4), C(4,4), L(4,4)
C
C DC 1 I=1,N
C DO 1 J=1,L
C 1 T(I,J) = 0.0
C
C DO 2 I=1,N
C DO 2 J=1,L
C 2 T(I,J) = 1, T(I,J) + A(I,K)*B(K,J)
C
C DO 3 I=1,N
C DO 3 J=1,L
C 3 C(I,J) = 1, T(I,J)
C
C RETURN
C
C END
C SUBROUTINE SUB (A,B,N,M,C)
C THIS SUBROUTINE SUBTRACTS THE NXM MATRIX B FROM THE NXM MATRIX
C A AND STORES THE RESULT IN C
C REAL*8 A, B, C
C DIMENSION A(4,4), B(4,4), C(4,4)

```

```

C      DC 1  I=1,N
C      DC 1  J=1,M
C      1 C(I,J) = A(I,J)-B(I,J)
C
C      RETURN
C
C      SUBROUTINE TRANS (A,N,M,C)
C      THIS SUBROUTINE FORMS THE MATRIX TRANSPOSE OF A STORING THE
C      RESULT IN C
C      A = NXM
C      C = MXN
C      REAL*8 A,C
C      DIMENSION A(4,4),C(4,4)
C
C      DO 1  I=1,N
C      DC 1  J=1,M
C      1 C(J,I) = A(I,J)
C
C      RETURN
C
C      SUBROUTINE YADD (X,Y,N,Z)
C      THIS SUBROUTINE COMPUTES THE SUM OF THE N-VECTORS X AND
C      Y AND STORES THE RESULT IN THE N-VECTOR Z
C
C      REAL*4 X(4),Y(4),Z(4)
C
C      DO 1  I=1,N
C      1 Z(I) = X(I)+Y(I)
C
C      RETURN
C
C      SUBROUTINE VPROD (A,X,M,N,Y)
C      THIS SUBROUTINE COMPUTES THE PRODUCT OF THE MXN MATRIX
C      A AND THE N-VECTOR X AND STORES THE RESULT IN THE
C      M-VECTOR Y
C
C      REAL*4 A(4,4),X(4),Y(4),T(4)
C
C      DO 1  I=1,M
C      T(I) = 0.D0
C      DO 1  J=1,N
C      1 T(I) = T(I)+A(I,J)*X(J)
C
C

```

2  $y(1)^2 - 1/M$ ,

CC RETURN  
END  
SUBROUTINE VREAD (V,N)  
CC THIS SUBROUTINE READS THE N-DIMENSIONAL S.P. VECTOR V  
DIMENSION V(4)  
READ (5,1) (V(I), I=1,N)  
RETURN  
FORMAT (8F10.0)  
1 END  
SUBROUTINE VSUB (X,Y,Z)  
REAL\*4 X(4), Y(4), Z(4)  
DO 1 I=1,N  
1 Z(I) = X(I) - Y(I)  
1 RETURN  
END  
SUBROUTINE VWRITE (V,N)  
DIMENSION V(4)  
WRITE (6,1) (V(I), I=1,N)  
RETURN  
FORMAT (9(2X,1PE12.5))  
1 END  
SUBROUTINE TRACK  
IF TRACK IS TO BE GENERATED ON-LINE IT IS DCNE IN THIS SUBROUTINE  
IN THE DEFAULT OPTION (ITRK=EQ.0) THE TRACK IS GENERATED  
FROM THE STANDARD LINEAR DIFFERENCE EQUATION  
X(K+1) = PHI\*X(K) + GAMMA\*W(K)  
REAL\*8 GAMMA, COVW, R, PHI, TEMP, TEMP1, PKK1, PKK2, Q, PKK, Q, EI, PR  
CC MMON EI(4,4), Q(4,4), PHI(4,4), TEMP(4,4), PKK(4,4), GAMMA(4,4), COVW(4,4)  
1 TEMP(4,4), TEMP1(4,4), TEMP2(4,4), H(4,4), PKK1(4,4), R(4,4), PHI(4,4),  
2 VAR(4,4,6,0), GKS(4,4,6,0), PKK2(4,4,6,0), XM(4,4,6,0), ERR(4,4,6,0)  
3 GAMMAS(4,4), PHI(4,4), X(4,4), HS(4,4), GK(4,4), SIGW(4,4), X(4,4)  
4 SIGWX(4,4), XZMEAN(4), XHKK(4), XHKK1(4), VTMP(4,2,4), V(4), SIGV(4),  
5 XHATZ(4,4), XZ(6,0), PY(10), PR(4,4)  
6 NNSAM1, Q, M1, ITR, ISTAT, K, ITRO, IXZ, IV, IW, ITEST, ND, NK  
DIMENSION W(3)  
TO GENERATE A SINGLE TRAJECTORY AND STORE IT IN THE ARRAY  
XS(1,K) I=1,N, K=2,NSAM (NOTE THAT IF A SINGLE TRAJECTORY IS TO BE  
GENERATED, THE INITIAL CONDITION HAS BEEN READ IN AND STORED  
IN XS(1,1), I=1,N)  
CC CCCCCC











```

4 SIGXZ(4) XZMEAN(4) XHKKM1(4) VTMP(4) Z(4),V(4),SIGN(4),
5 XHATZ(4) XZ(60) YZ(60) PY(10) PR(4)4
6N NSAM(1Q, M1TER, ITRK, IN, ISTAT, K, ITRO, IXZ, IV, IW, IEST, ND, NR
WRITE(6,147)

C WRITE GAINS, THEORETICAL COVARIANCES OF ESTIMATION ERROR
C WRITE(6,148)
C DC 59 K=1,NSAM
C WRITE(6,149) K
C DO 59 I=1,N
C WRITE(6,146) (GKS(I,J,K),J=1,M)
C WRITE(6,150)
C DO 60 K=1,NSAM
C WRITE(6,151) K
C DO 60 I=1,N
C WRITE(6,146) (PKKS(I,J,K),J=1,N)
C
C 61 WRITE(6,156)
C WRITE(6,152)
C WRITE(6,153)
C DO 62 K=1,NSAM
C WRITE(6,155)
C DO 62 I=1,N
C WRITE(6,154) K, I, XM(1,I,K), ERR(1,I,K), VAR(1,I,K)
C
C 61 WRITE(6,156)
C WRITE(6,152)
C WRITE(6,153)
C DO 62 K=1,NSAM
C WRITE(6,155)
C
C 62 WRITE(6,156)
C
C 146 FORMAT(9(2X,1PE12.5)')/
C 147 FORMAT(10X,'THE GAIN MATRICES ARE: //')
C 148 FORMAT(5X,'K= 13 / 10X.G(K)= /')
C 149 FORMAT(1X,'/ 10X. THE THEORETICAL COVARIANCE MATRIX IS: //')
C 150 FORMAT(5X,'K= 13 / 10X.P(K/K)= /')
C 151 FORMAT(5X,'K= 13 / 10X. VECT. MEAN',/1)
C 152 FORMAT(5X,'TIME',/16, 'VECTOR COM- /')
C 1 T51 'SAMPLE MEAN OF', T71 'SAMPLE VARIANCE CF',/1
C 153 FORMAT(5X,'INDEX', T16 'PONENT INDEX', T34 'CF TRACK',/1
C 1 T51 'ESTIMATION ERROR', T71 'ESTIMATION ERROR',/1

```

```

154 FORMAT (6X,13,13X,11,10X,1PE14.7,2(6X,1PE14.7))
155 FORMAT (//,1.)
156 FORMAT (10X, THE SAMPLE COVARIANCE OF EST. ERROR MATRIX IS*,//)
157 FORMAT (//,2X, K=*,13,/)
158 FORMAT
      RETURN
END
SUBROUTINE PLT
REAL*8 GAMMA, COVW, R, PHI, H, TEMP1, TEMP2, PKKM1, G, PKK, Q, EI, PR
COMMON EI(4,4), Q(4,4), PKK(4,4), GAMMA(4,4), COVW(4,4), R(4,4), PHI(4,4)
TEMP(4,4), TEMP1(4,4), TEMP2(4,4), H(4,4), PKKM1(4,4), R(4,4), PHI(4,4),
2VAR(4,4,60), GKS(4,4,60), PKS(4,4,60), XM(4,4,60), ERR(4,4,60), CH200083
3GAMMAS(4,4), PHIS(4,4), XS(4,4,60), HS(4,4,60), GK(4,4), SIGW(4,4),
SIGGXZ(4,4), XZMEAN(4), XHKK(4), VTM(4,4), X(4), V(4), SIGV(4),
5XHATZ(4,4), XZ(60), YZ(60), PX(10), PY(10), PR(4,4,4),
6N1NSAM, 1Q, MITER1, ITRK1N, ISTAT, K, ITRO, IXZ, IV, IW, IEST, ND, NR
INTEGER*4 ITB(12)/12*0/
REAL*4 RTB(28)/28*0.0/
DIMENSION XP(60), YP(60)
EQUIVALENCE (TITLE, RTB(5))
REAL*8 TITLE(12), X = TRUE, + = FILTER, SQUARE = NOISY*/

C
1GP LT=1
1THVPL=1
1WTPL=1
1SVPLT=1
1SVPLT=1
DO 500 KY=1, NR
  KX=NR+1-KY
  DO 50 K=1, NSAM
    XP(K) = XM(KX, 1, K)
    YP(K) = XM(KX, 3, K)
    CALL PLOT(XP, YP, NSAM, 0)
  500 CONTINUE
  ITB(1)=1
  ITB(2)=1
  CALL DRAWP(60, XP, YP, ITB, RTB)
  DO 51 K=1, NSAM
    XP(K) = XS(1, 1, K) + ERR(1, 1, K)
    YP(K) = XS(1, 3, K) + ERR(1, 3, K)
  51 CALL PLOT(XP, YP, NSAM, 0)
  ITB(1)=2
  ITB(2)=2
  CALL DRAWP(60, XP, YP, ITB, RTB)
  DO 2 J=1, 605
    IF (ABS(PKK(1,1, J) - PKKS(3, 3, J)) .GT. 0) GO TO 11

```

```

11  PKKS(1,1,J)=PKKS(3,3,J)+0.000001
    CONTINUE
    TFE=0.5*TAN(2.*PKKS(1,3,J)/(PKKS(1,1,J)-PKKS(3,3,J)))
    IF(ABS(TFE).GT.0) GO TO 10
    TFE=0.00001

10  CONTINUE
    SIG2X=(PKKS(1,1,J)+PKKS(3,3,J))/2.+PKKS(1,3,J)/SIN(2.*THE)
    SIG2Y=(PKKS(1,1,J)+PKKS(3,3,J))/2.-PKKS(1,3,J)/SIN(2.*THE)
    WRITE(6,146)
    146 FORMAT(9(2X,1PE12.5),/)

    SX=(SIG2X)**.5*20.
    SY=(SIG2Y)**.5*20.
    PT=3.14159265/12.
    CT=COS(THE)
    ST=SIN(THE)
    DC1=1,I=1,25
    XI=1
    XP(1)=SX*COS(PT*X1)*CT-SY*SIN(PT*X1)*ST+XS(1,1,J)
    YP(1)=SX*COS(PT*X1)*ST+SY*SIN(PT*X1)*CT+XS(1,3,J)
    1  CALL DRAWP(25,XP,YP,ITB,RTB)
    DO 201 J=2,NR
    DO 200 K=1,NSAM
    DO 200 K=1,NSAM
    XP(K)=XS(J,1,K)
    YP(K)=XS(J,3,K)
    IT=(J/2)+3
    ITB(2)=IT
    201 CALL DRAWP(60,XP,YP,ITB,RTB)
    ITB(1)=3
    ITB(2)=3
    CALL DRAWP(60,XZ,YZ,ITB,RTB)
    DO 65 K=1,NSAM
    65 XP(K)=K
    C   IF (IGPLT.NE.1) GO TO 68
    C   DO 67 I=1,N
    C   DO 67 J=1,M
    C   DO 66 K=1,NSAM
    C   66 YP(K)=GKS(1,J,K)
    C   WRITE(6,156)
    CALL PLOT(XP,YP,NSAM,0)
    67 WRITE(6,159) I,J
    C   68 IF (ITHVPL.NE.1) GO TO 71
    C

```

```

      DO 70 I=1,N
C      DO 69 K=1,NSAM
C      69 YP(K) = PKS(I,I,K)
C      WRITE (6,156)
C      CALL PLOT (XP,YP,NSAM,0)
C      70 WRITE (6,160) I,I
C      71 IF (IMPLT.NE.1) GO TO 74
C      DO 73 I=1,N
C      DO 72 K=1,NSAM
C      72 YP(K) = XM(I,I,K)
C      WRITE (6,156)
C      CALL PLOT (XP,YP,NSAM,0)
C      73 WRITE (6,161) I,I
C      74 IF (ISMPLT.NE.1) GO TO 77
C      DO 76 I=1,N
C      DO 75 K=1,NSAM
C      75 YP(K) = ERR(I,I,K)
C      WRITE (6,156)
C      CALL PLOT (XP,YP,NSAM,0)
C      76 WRITE (6,162) I,I
C      77 IF (ISVPLT.NE.1) GO TO 80
C      DO 79 I=1,N
C      DO 78 K=1,NSAM
C      78 YP(K) = VAR(I,I,K)
C      WRITE (6,156)
C      CALL PLOT (XP,YP,NSAM,0)
C      79 WRITE (6,163) I,I
C      80 CONTINUE
C      156 FORMAT (1,1)
C      159 FORMAT (12X,6(1,1),12X,6(1,1),VS,K)
C      160 FORMAT (12X,PKK(1,1,1,1,1,1),VS,K)
C      161 FORMAT (12X,MEAN OF X(1,1,1,1,1,1),VS,K)

```

162 FORMAT (12X,'XHATKK('),11,'-X('),11,') VS.., K')
 163 FORMAT (12X,'ERROR VARIANCE('),11,') VS.., K')
 RETURN
 END
 //GO.FT06FO01 DD SYSOUT=A,SPACE=(CYL,(4,1))
 //GO.SYSIN DD \*
 \$\$\$\$\$\$\$\$\$DATA DECK\$\$\$\$\$\$\$\$\$  
 ND 4 N 4 NSAM 60 NENS 1 NR 4
 IPRT 1 IPLT 1 PHI 1.0 1.0 1.0 1.0
 1.0 1.0 1.0 1.0 1.0 1.0 1.0 1.0
 1.0 R 0.0001 COVW 0.0001
 0.0001 GAMMA 0.0001
 0.5 1.0 0.5 1.0 0.5 1.0 0.5 1.0
 PRR(1) 0.0001 0.0001 0.0001
 0.0001 0.0001 0.0001 0.0001
 PRR(2) 0.0001 0.0001 0.0001
 0.0001 0.0001 0.0001 0.0001
 PRR(3)

•0001      •0001      •0001  
•0001      •0001      •0001  
SIGV

XHATZ - JET      10.  
0.3333      XHATZ - INFANTRY1  
XHATZ - INFANTRY2      10.  
XHATZ - HELICOPTER  
-•05555555      XS - JET      10.  
0.3333      XS - INFANTRY1  
XS - INFANTRY2      10.  
XS - HELICOPTER      10.  
-•05555555      12.

0.00166667  
-0.00166667

10.  
12.

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